

APPLICATION UNDER UNITED STATES PATENT LAWS

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Invention: FLUID FLOW MACHINE (TURBOMACHINE) WITH INCREASED ROTOR-
STATOR RATIO

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This is a:

- ☐ Provisional Application
- ☒ Regular Utility Application
- ☐ Continuing Application
_____ The contents of the parent are
incorporated by reference
- ☐ PCT National Phase Application
- ☐ Design Application
- ☐ Reissue Application
- ☐ Plant Application

This application claims priority to German Patent Application No. DE 10233033.6, filed July
20, 2002 which application is incorporated by reference herein.

SPECIFICATION

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5 incorporated by reference herein.

BACKGROUND OF THE INVENTION

This invention relates to a fluid-flow machine (turbomachine). More specifically, this invention relates to a fluid-flow
10 machine (turbomachine) with at least one rotor equipped with blades and with at least one stator equipped with vanes, with the rotor being supported in a casing by means of a rotating shaft.

15 The aerodynamic loading of fluid-flow machines (turbomachines), such as fans, compressors, pumps and blowers, which can be operated with gaseous and liquid media, is determined by the growth and the separation of boundary layers both on the blades and on the hub as well as the casing walls. Normally, good
20 operating characteristics are obtained if the aerodynamic load is distributed uniformly on the rotor blades and the stator vanes, for example in a fan or compressor stage. The annulus contraction of the rotor and the stator, i.e. the decrease of the cross-sectional area of the annulus in the direction of flow
25 necessary for a compressive working media between the entry and exit plane of a stage, is here very similar. Minor differences in the contraction of the rotor and the stator result from the degree of reaction selected for the respective stage.

In designs according to the state of the art, the inner and outer annulus contours, for example of fans and compressors, converge continuously showing tapering without unevenness.

5 Accordingly, the annuli known from the state of the art fall into two categories, the first one being annuli which show a continuous taper and the second one being annuli which show a taper, but have a constant area in the axial gaps between the blades/vanes, with no contraction being present. The latter
10 state of the art becomes apparent from US Patent 6,312,221 B1, for example.

BRIEF SUMMARY OF THE INVENTION

15 In a broad aspect, the present invention provides a fluid-flow machine (turbomachine) of the type specified at the beginning which, while being simply designed and dependable in operation, will aerodynamically unload the rotor.

20 It is a particular object of the present invention to provide a solution to the above problem by the features described herein, with further objects and advantages of the present invention becoming apparent from the description below.

25 The present invention accordingly provides a form of annulus whose cross-sectional area in at least one stage consisting of one rotor and one stator results in a rotor-stator contraction ratio QRS which satisfies the following equation:

30
$$[0.2 + (KT - 0.45)^{0.1}] < QRS < 3.0$$

where KT is the total-stage contraction. QRS and KT are calculated as follows:

$$QRS = KR/KS \text{ with } KR = ARI/ARA$$

$$\text{and } KS = ASI/ASA$$

$$KT = ARI/ASA$$

where ARI , ARA , ASI and ASA are calculated as follows:

$$ARI = \pi (R_2^2 - R_1^2)$$

$$ARA = \pi (R_4^2 - R_3^2)$$

$$ASI = \pi (R_6^2 - R_5^2)$$

$$ASA = \pi (R_8^2 - R_7^2)$$

where in the direction of flow of the fluid-flow machine

R_1 is the radius at the base point of the rotor blade leading edge on the rotor shaft,

R_2 is the radius at the radial outer point of the rotor blade leading edge,

R_3 is the radius at the base point of the rotor blade trailing edge on the rotor shaft,

R_4 is the radius at the radial outer point of the rotor blade trailing edge,

R_5 is the radius at the radial inner point of the stator vane leading edge,

R_6 is the radius at the radial outer point of the stator vane leading edge,

R_7 is the radius at the radial inner point of the stator vane trailing edge, and

R_8 is the radius at the radial outer point of the stator vane trailing edge.

In an alternative form, the present invention provides a form of annulus whose cross-sectional areas result in axial-gap contractions $KX1$ and $KX2$ which satisfy the following equations:

$$0.8 < KX1 < 1.0$$

$$0.8 < KX2 < 1.0$$

where $KX1$ and $KX2$ satisfy the following equations:

$$KX1 = ARA/ASI$$

$$KX2 = ASA/ARI2$$

where ARA , ASI , ASA and $ARI2$ are calculated as follows:

$$ARA = \pi (R_4^2 - R_3^2)$$

$$ASI = \pi (R_6^2 - R_5^2)$$

$$ASA = \pi (R_8^2 - R_7^2)$$

$$ARI2 = \pi (R_{10}^2 - R_9^2)$$

where, in the direction of flow of the fluid-flow machine (turbomachine),

- R_3 is the radius at the base point of the rotor blade trailing edge on the rotor shaft,
- 5 R_4 is the radius at the radial outer point of the rotor blade trailing edge,
- R_5 is the radius at the radial inner point of the stator vane leading edge,
- 10 R_6 is the radius at the radial outer point of the stator vane leading edge,
- R_7 is the radius at the radial inner point of the stator vane trailing edge,
- 15 R_8 is the radius at the radial outer point of the stator vane trailing edge,
- R_9 is the radius at the radial inner point of the leading edge
20 of the downstream next rotor blade, and
- R_{10} is the radius at the radial outer point of the leading edge of the downstream next rotor blade.

25 The present invention is characterized by a variety of merits.

The invention relates to a fluid-flow machine (turbomachine) of axial and semi-axial design for both gaseous and liquid working media.

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The solution according to the present invention is useful for fluid-flow machines (turbomachines) with only one stage (rotor and stator) and with multiple stages.

5 As already known from the state of the art, the rotor comprises a number of blades which are fitted to the rotating shaft of the fluid-flow machine and transfer energy to the working medium. Accordingly, the stator consists of a number of stationary vanes which are either shrouded on both ends or have a free end at the
10 side of the hub. Both, rotor and stator, are situated in a casing.

Furthermore, the fluid-flow machine in the form according to the present invention can be provided with a stator (inlet guide
15 vanes) in front of the first rotor. The vanes of the stators can be rotatable, enabling them to be adjusted from outside of the casing, for example by means of a jackscrew-type actuator.

In the case of the first solution according to the present
20 invention which relates to the contraction ratio Q_{RS} between the rotor and the stator, it is particularly favourable if the fluid-flow machine, which, for example, can be a turbomachine, comprises stators with special high-lift configuration and correspondingly high aerodynamic loadability. Under this aspect,
25 it is highly advantageous to provide the individual stages of this fluid-flow machine (turbomachine) with a form of annulus whose cross-sectional area, while holding the rotor loading constant, enables the stator loading to be increased significantly. Different from the state of the art, it is in
30 this case useful to provide a notable excess of the rotor-stator contraction ratio of a stage. Considering a stage of a multi-stage fluid-flow machine (turbomachine), the contraction at the

rotor in the stage is relatively high, while the contraction at the stator in the same stage is much lower. This situation can be quantified by the contraction ratio QRS between rotor and stator according to the present invention.

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BRIEF DESCRIPTION OF THE DRAWINGS

Fig. 1 shows a section of a multi-stage turbomachine with a definition of the cross-sectional areas of the annulus;

10 Fig. 2 shows the formulas for the calculation of the five cross-sectional areas of the annulus;

Fig. 3 shows an example of a form of annulus according to the present invention in comparison with an annulus according to the
15 state of the art; and

Fig. 4 shows an example of an annulus form according to the present invention in comparison with an annulus according to the state of the art.

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DETAILED DESCRIPTION OF THE INVENTION

This detailed description should be read in conjunction with the brief summary of the invention above.

25 Fig. 1 shows the n-th stage of a multi-stage turbomachine. The four corner points of the rotor (n) and of the stator (n) as well as the forward corner points of the subsequent rotor (n+1) are specially marked. The corner points are defined by the sections of the leading and trailing edges (or their radial
30 extensions in the case of rotor blade or stator vane ends with an annular gap) with the casing or the hub, respectively. If the

hub or casing contour in the circumferential direction does not extend with a constant radius, the arithmetically averaged radius in the circumferential direction will apply. The rotor (n) has the points P1 to P4, the stator (n) has the points P5 to P8, and the rotor (n+1) has the points P9 to P10. Each point P(i) has the axial position X(i) and the Radius R(i). The coordinate reference system lies on the machine axis.

Fig. 2 shows the formulas for the calculation of the five cross-sectional areas of the annulus ARI, ARA, ASI, ASA and ARI2 that can be calculated from the radii. Accordingly, the rotor contraction is obtained from $KR = ARI/ARA$, while the stator contraction is calculated from $KS = ASI/ASA$.

The contraction ratio between rotor and stator thus is $QRS = KR/KS$.

Furthermore, the total-stage contraction $KT = ARI/ASA$ is important.

The present invention relates to a range of increased values of QRS in dependence of KT according to the following rule:

$$[0.2 + (KT - 0.45)^{0.1}] < QRS < 3.0$$

The relationship according to the above formula is shown in graphical form on the right-hand side of Fig. 2. As is apparent, the fluid-flow machines (turbomachines) according to the state of the art lie clearly below the characteristic.

Fig. 3 shows an example of a form of annulus according to the present invention in comparison with an annulus according to the

state of the art. Using this annulus configuration according to the state of the art as the basis, the QRS criterion can be satisfied by solely the shape of the hub contour of the rotor shaft, by solely the shape of the casing contour, or by a combination of shapes of the hub and the casing contours, respectively.

Obviously, a highly simplified form of representation has been selected in Fig. 3. It would also be possible to change the rotor entry and exit points.

The above alternative of the solution in accordance with the present invention takes into account the extension of the static pressure build-up, which is normally confined to the rotor and stator rows, to the axial gaps between the individual blade/vane rows. This situation can also be described in quantitative terms using the cross-sectional areas of the annulus at the entry and exit planes of the rotors and stators. For this purpose, reference is again made to Fig. 1.

As shown in Fig. 2, the given cross-sectional areas of the annulus ARA, ASI, ASA and ARI2 permit the calculation of the contraction $KX1 = ARA/ASI$ between rotor and stator and of the contraction $KX2 = ASA/ARI2$ between stator and subsequent rotor. Accordingly, the invention provides for a range of values of KX below 1.0. Here, the following rules apply:

$$0.8 < KX1 < 1.0 \text{ and} \\ 0.8 < KX2 < 1.0.$$

The values known from the state of art always exceed 1.0 or amount to 1.0, respectively.

Fig. 4 shows an example of an annulus form according to the present invention in comparison with an annulus according to the state of the art. Using a given annulus configuration (state of the art) as the basis, the KX criterion according to the present invention can be satisfied by solely the shape of the hub contour, by solely the shape of the casing contour, or by a combination of the shapes of the hub and the casing contours, respectively.

For reasons of clarity, the possibility to change rotor and stator exit points was also dispensed with in Fig. 4.

The comparison with the state of the art reveals that the annuli known from the state of the art do not include the use of highly different contractions (area ratios) in rotor and stator. Accordingly, the solution according to the present invention is highly advantageous if different design methodologies are applied for rotors and stators. These may comprise the use of traditional techniques at the rotor and the application of high-lift configurations at the stator. All fan and compressor designs according to the state of art show well-defined differences in the percentage decrease of the annulus areas in the direction of flow across the rotor and the stator of one stage. Also, the state of the art does not take account of any partial relocation of diffusion from the blade/vane rows to the axial gap. This means that a decreasing or constant cross-sectional area of the annulus in the direction of flow is assumed in the state of the art.

Therefore, it is a particular advantage of the present invention that the use of an increased rotor-stator contraction ratio

creates the prerequisites for the design of a fluid-flow machine, or a turbomachine, which includes the hybrid application of a conventionally developing rotor technology and a novel technology for stators in high-lift configurations. The approach according to the present invention is, therefore, completely novel and different from the state of the art, creating a distinguished category of new fluid-flow machines or turbomachines.

10 For a given pressure ratio of a turbomachine, the number of components installed can be reduced by approx. 40 percent, with efficiency being constant or improved (by up to 2 percent depending on the operation of the special stators mentioned). The resultant cost saving is approx. 20 percent.

15 If this concept is used in the compressor of an aircraft engine with approx. 25,000 lbs thrust, specific fuel consumption will be reduced by up to 1 percent.

20 Furthermore, the application of axial-gap contractions below 1.0 presents an opportunity to include the axial gaps, which normally are unused, as active elements in the static pressure build-up process within a fluid-flow machine (turbomachine), thus utilizing more effectively a given axial build length. This approach is also completely novel with regard to its overall concept and is fully left out of consideration in the state of the art. This approach in itself creates a distinguished category of novel fluid-flow machines (turbomachines). For a given pressure ratio of the fluid-flow machine (turbomachine), efficiency can be increased by approx. 0.5 percent and the number of blades/vanes installed reduced by 5 percent. The resultant cost saving amounts to approx. 2 percent. If the

- concept is used in the compressor of an aircraft engine with approx. 25,000 lbs. thrust, specific fuel consumption is reduced by approx. 0.25 percent.